Amendments to the Claims:

This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims:

Claims 1-5. (Cancelled)

6. (Currently Amended) A rotor for a gas turbine, having blades arranged on the rotor and rotating together with the rotor, the blades forming a blade ring and the blades within the blade ring being arranged at a different distance from one another and thus with a different blade pitch, a distance between the blades within the blade ring changing continuously or discontinuously in a circumferential direction and the distance between the blades within the blade ring dimensioned such that an imbalance is canceled out, wherein the rotor has several rotatable blade rings arranged axially behind one another, and wherein, within each blade ring, the blades are arranged at a different distance from one another.

7. (Original) The rotor according to Claim 6, wherein an arrangement of blades within a respective blade ring is different for each of the blade rings.

8. (Original) The rotor according to Claim 6, wherein the rotor is a turbine rotor or compressor rotor of a gas turbine.

9. (Original) The rotor according to Claim 8, wherein the gas turbine is a turbine of an airplane engine.

10. (Original) The rotor according to Claim 6, wherein the rotor is a fan rotor of a gas turbine.

11. (Original) The rotor according to Claim 10, wherein the gas turbine is a turbine of an airplane engine.

12. (Original) The rotor according to Claim 6, wherein the rotor is a blisk (bladed disk) or bling (bladed ring) of a gas turbine and wherein the blades are an integral component of the rotor.

13. (Original) The rotor according to Claim 12, wherein the gas turbine is a turbine of an airplane engine.

14. (Currently Amended) A rotor for a gas turbine, comprising:

a first rotatable blade ring; and

a second rotatable blade ring;

wherein the first and the second rotatable blade rings are arranged axially one behind the other and wherein on each rotatable blade ring a first blade is positioned at a first distance from a second blade and the second blade is positioned at a second distance from a third blade, wherein the first distance is different from the second distance.

15. (Original) The rotor according to Claim 14, wherein on each blade ring a fourth blade is positioned at the first distance from a fifth blade and the fifth blade is positioned at the second distance from a sixth blade and wherein the first, second, and third blades are located diametrically opposed from the fourth, fifth, and sixth blades, respectively, on the blade ring.

16. (Original) The rotor according to Claim 14, wherein an arrangement of blades within a respective blade ring is different for each of the blade rings.

17. (Currently Amended) A method for optimizing vibrations in a gas turbine engine, comprising the steps of:

rotating a first blade ring of a rotor;

rotating a second blade ring of ~~a~~ the rotor, wherein the first and second rotating blade rings are arranged axially behind one another;

wherein on each blade ring a first blade is positioned at a first distance from a second blade and the second blade is positioned at a second distance from a third blade and wherein the first distance is different from the second distance; and

continuously changing a frequency of a vibration of a stationary assembly of the gas turbine by rotating the first and second blade rings.